

B1
nozzle guide vane annular. The inner surfaces (ie those facing into the gas flow 3) of the vane platforms are smooth-flow walls.

Page 5, line 21-page 6, line 2, delete current paragraph and insert therefor:

B2
The circumferential flange 40 is provided with a series of axial apertures 76, each in approximate axial alignment with a corresponding aperture 68 in the shroud liner segment 24, thus enabling relatively cool high pressure compressor air to pass from the annular volume 19 through the apertures 68 into the interior of the box liners. This air then exits the interior of box liner segment 24 through aperture(s) 74 into the inter-liner gaps 78. The cross-sections of apertures 76 and 68 will be chosen so that despite the radial position of the shroud liner segment 24 there will be a sufficient overlap between the apertures 76 and 68 for high pressure compressor air to flow therethrough. The rate at which air exits the box liners is determined, ie metered, by the exit apertures 74.

IN THE CLAIMS:

Please amend the claims as follows:

Please replace claims 1 and 8 as follows:

B3
1. (Twice Amended) A pressure actuated tip clearance system for a shroud structure of a gas turbine rotary stage including an annular plenum chamber formed between an annular arrangement of a plurality of shroud liners on the inner circumference of the chamber and a generally cylindrical casing on the radially outer side, and, in use, a hot gas stream located radially inwards of the shroud liners, wherein each shroud liner comprises a hollow box section comprising upstream and downstream walls, radially inner and outer walls, and side walls, at least the downstream wall and the radially outer wall being closed, the upstream wall having an air inlet aperture, and at least one of the side walls having at least one